

COMPARISON OF SQUARE AND CIRCULAR SECTIONED WIND TUNNEL PERFORMANCE

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ABSTRACT

In this study, square and circular sectioned subsonic wind tunnel performance are examined numerically. The computational fluid dynamic (CFD) analysis of three-dimensional (3D) flow in square and circular sectioned wind tunnels are used for comparison. Experimental studies are very important in designing a new product. In aerodynamic designs, wind tunnel tests are commonly used in experimental studies. In experimental studies, accuracy is the most critical point to verify the results. Turbulence intensity is the main drawback of wind tunnels. Hence in this study, turbulence intensity of the square and circular sectioned desktop size wind tunnel is investigated.

Keywords: CFD, Wind tunnel, Turbulence

1. INTRODUCTION

In the design stage of a new product, numerical and experimental studies are commonly used. Numerical analyses should be performed for the new designs but these analyses do not give accurate results for some analyses types. Hence, numerical applications should be verified by experimental studies. Experimental studies are very important in designing a new product. In experimental studies, accuracy is the most critical point to verify the numerical results. Experimental setup directly affects the accuracy, to obtain more accurate results setup must be optimized. In aerodynamic designs, wind tunnel tests are commonly used in experimental studies. Turbulence intensity is the main drawback of wind tunnels. It must be minimized to improve the accuracy of the experiments.

In the literature, many different studies exist about the aerodynamic applications, wind tunnels, and flow effects on temperature and surfaces. The design and the aerodynamic and acoustic performance of wind tunnel applications can be found in the literature [1, 2, 3, 4, 5, and 6]. Many different aerodynamic experimental studies are performed with aerodynamics concepts and wind tunnels to verify the numerical results [7, 8, 9, 10, 11, 12, 13, 14, 15, 16, 17, and 18]. Also, many different experimental applications exist in the

literature about the flow effect on surfaces such as abrasion and erosion [19, 20, and 21].

2. ANALYSIS

In this section, the turbulence intensity of the square and circular sectioned desktop size wind tunnel is investigated. In the analyses, the SolidWorks Flow Simulation tool is used. Square and circular sectioned wind tunnel models (in figure 1) with 0.03 m^2 section area and 0.8 m length are used. Inlet flow rate is applied as $1.2 \text{ m}^3/\text{s}$. The element number of the finite model is 375000. Flow simulation results are given in 6 sections (in figure 2).

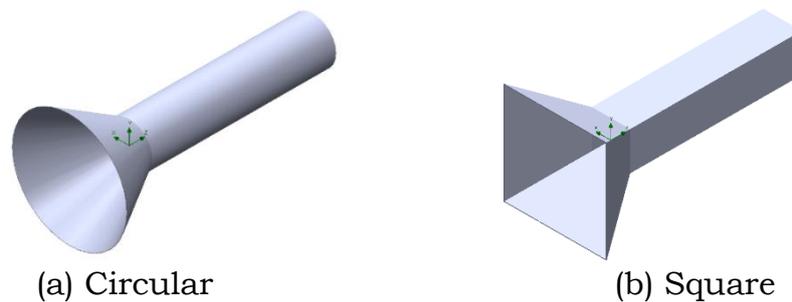


Figure 1. Solid models of Tunnels

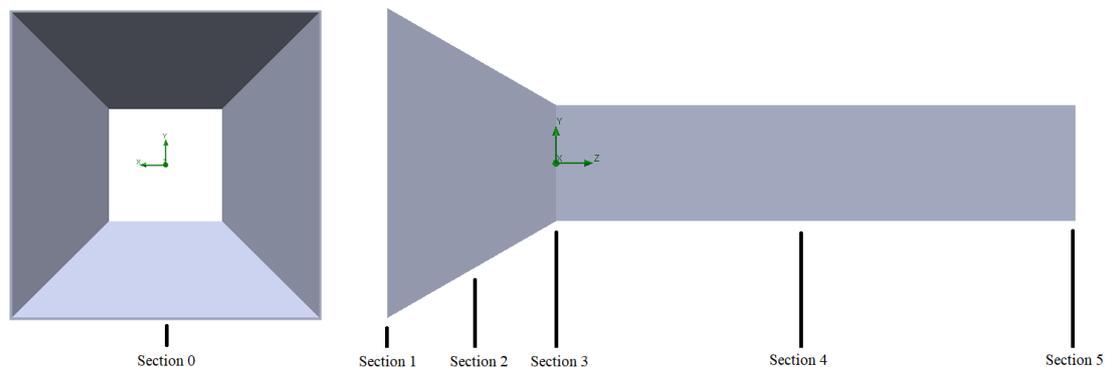
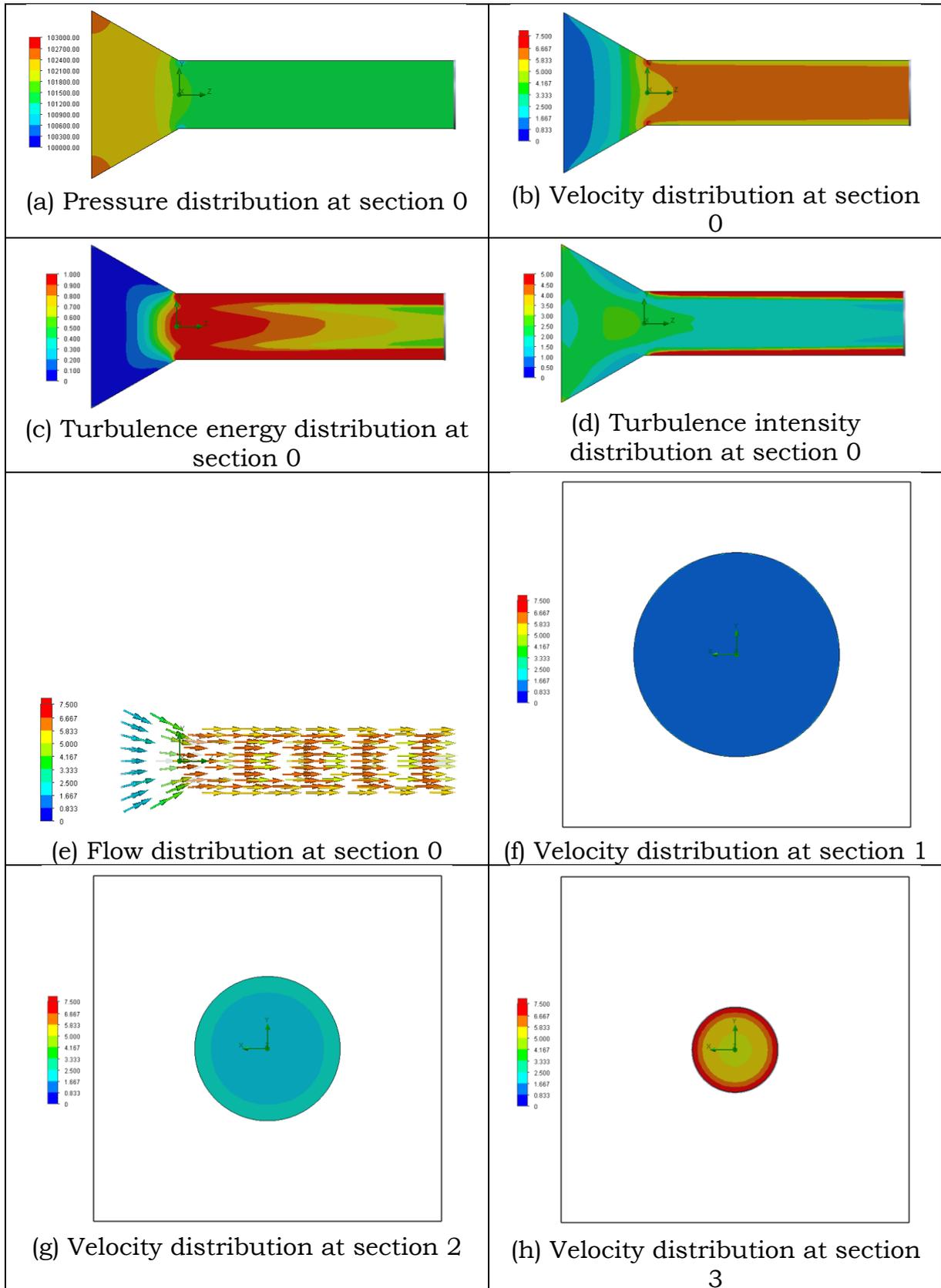


Figure 2. Sections for Results

3. ANALYSIS RESULTS

Two different desktop-size wind tunnel models are used in the analyses which are square and circular sectioned models. In the aerodynamic analyses, pressure distribution, velocity distribution, and turbulence intensity distribution are important parameters. Hence for the comparison, these parameters are used as performance parameters. Circular sectioned models results are given in figure 3 and square-sectioned models results are given in figure 4. In the literature, the test section is selected between the inlet and the fan. Hence, section 4 in figure 2 is defined as the test section.



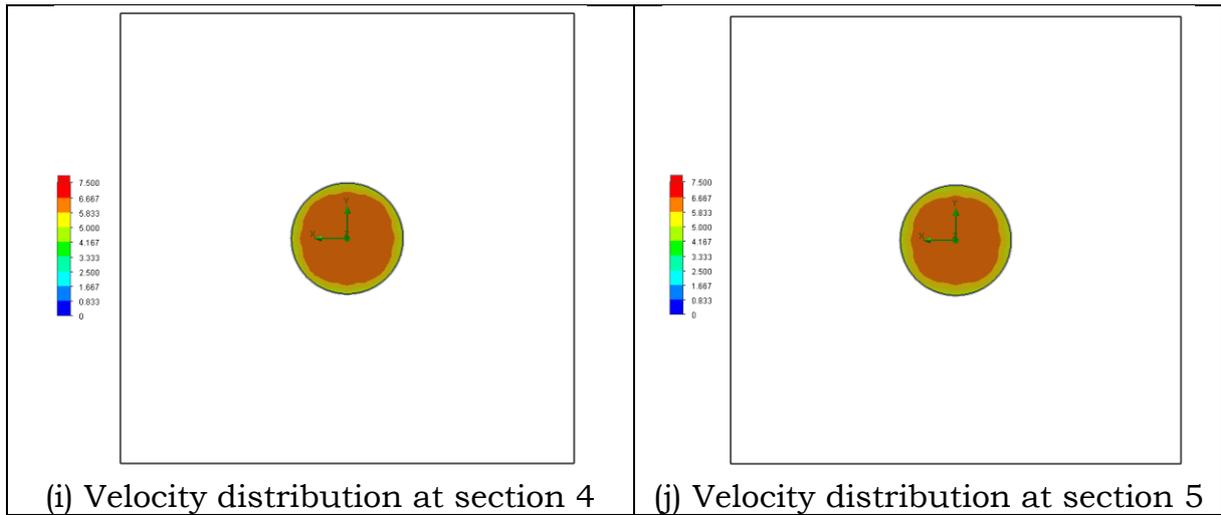
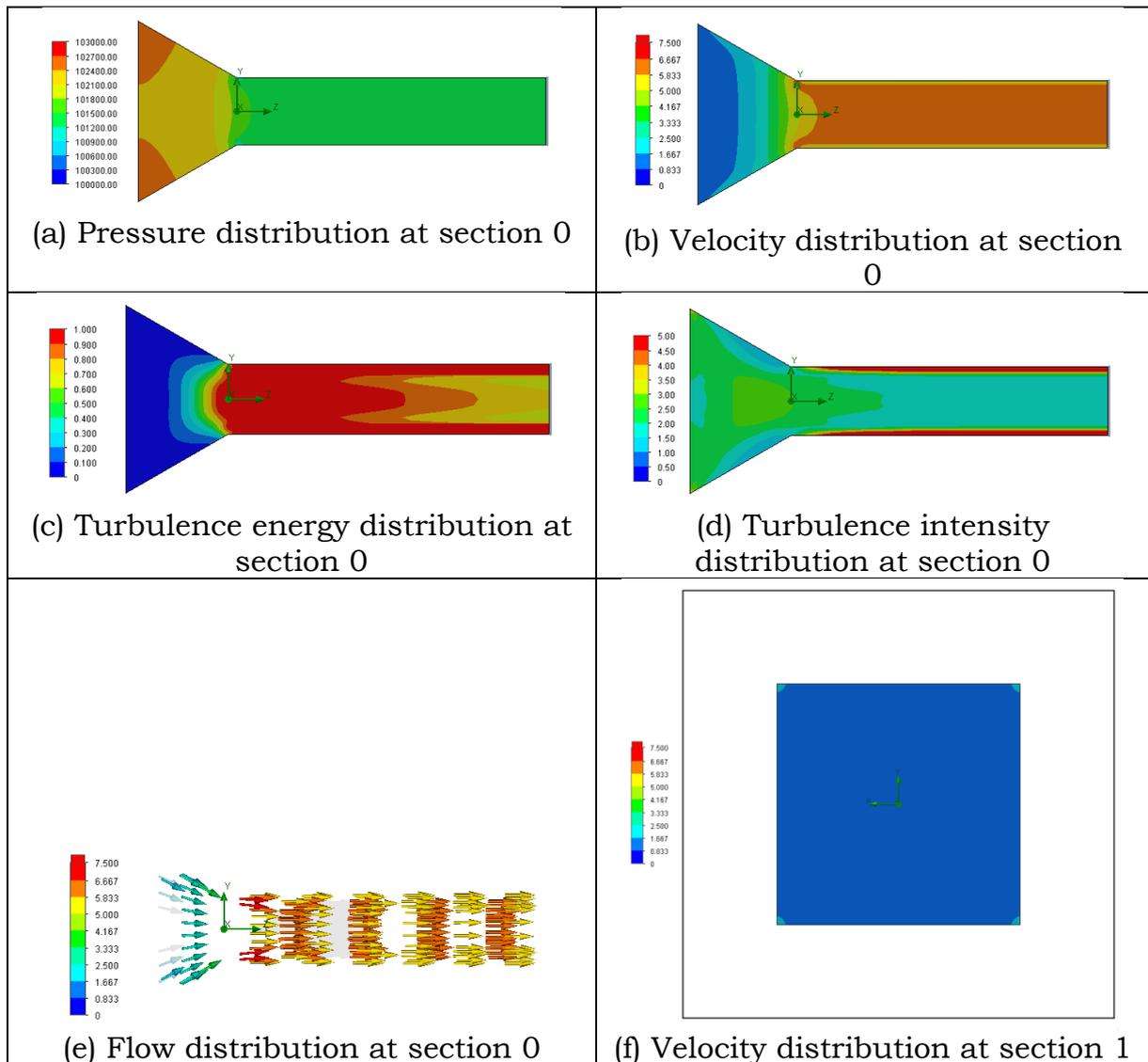


Figure 3. Circular sectioned wind tunnel model results



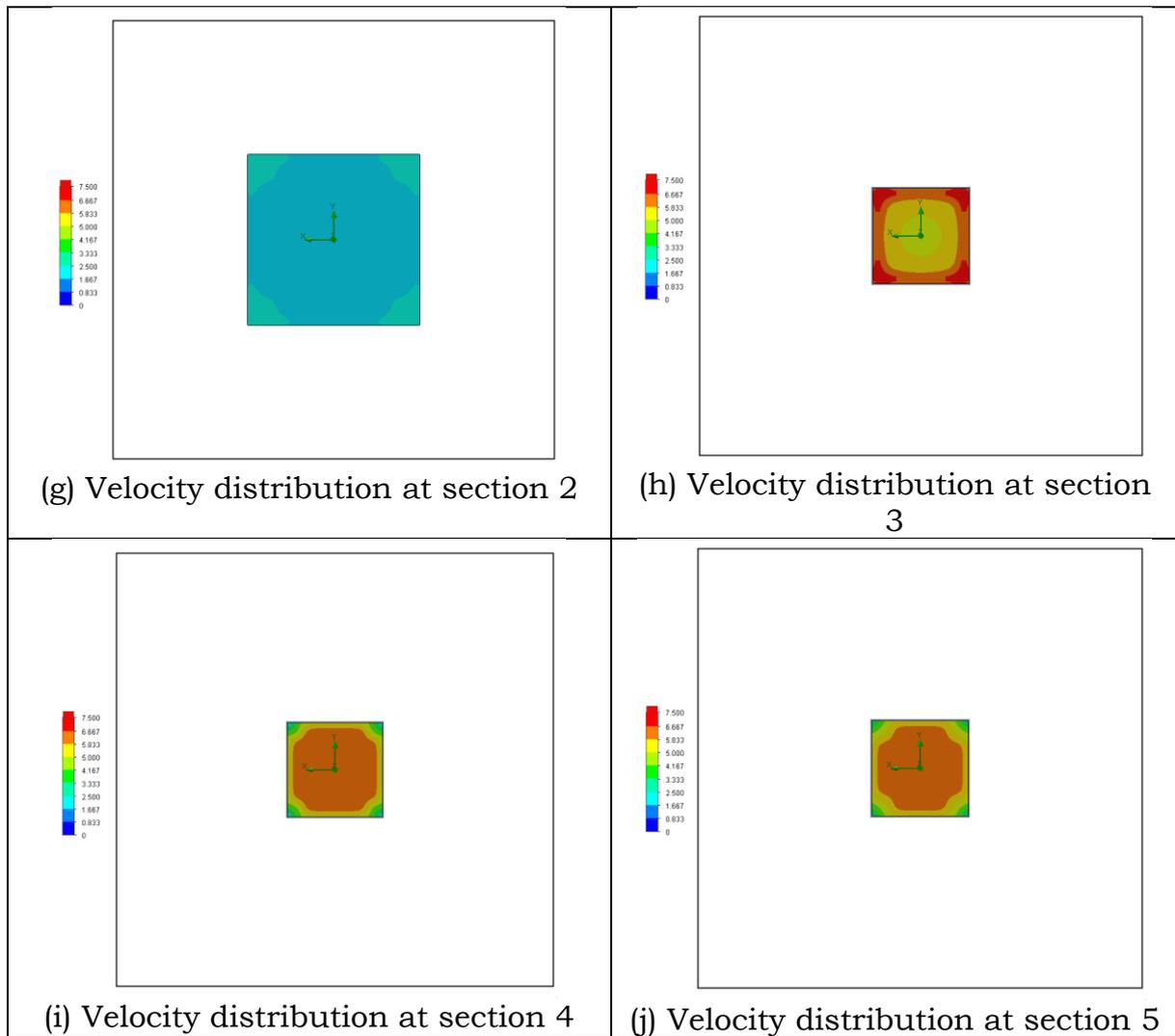


Figure 4. Square sectioned wind tunnel model results

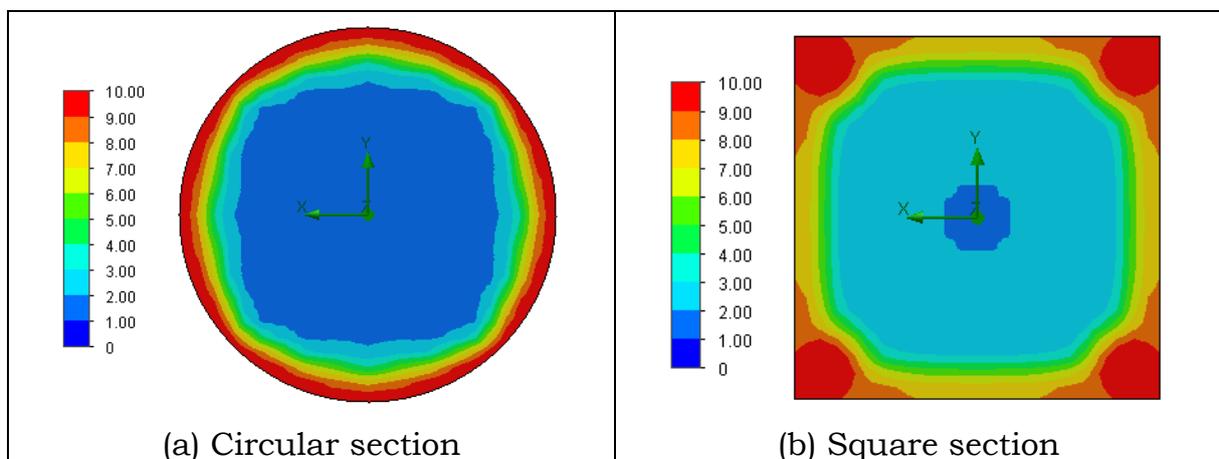


Figure 5. Turbulence intensity distribution at section 4

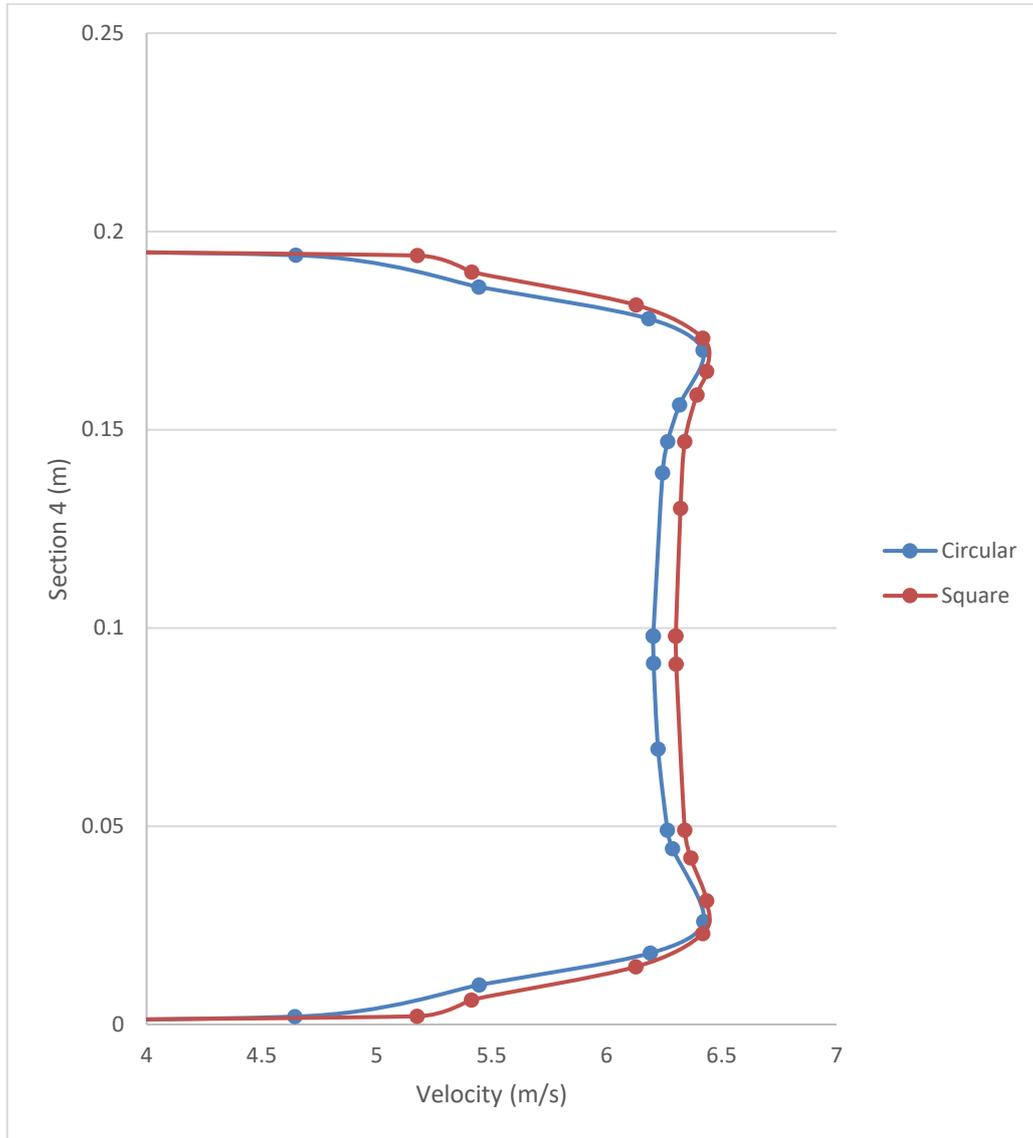


Figure 6. Velocity values at section 4

When the Figure 6 is examined, it is seen that when compared in terms of flow velocity, it has been determined that a velocity profile close to the tunnel entrance velocity is obtained in the circular cross-section wind tunnel compared to the square cross-section wind tunnel. The fact that the obtained speed value is close to the input speed increases the reliability of the experiments to be performed.

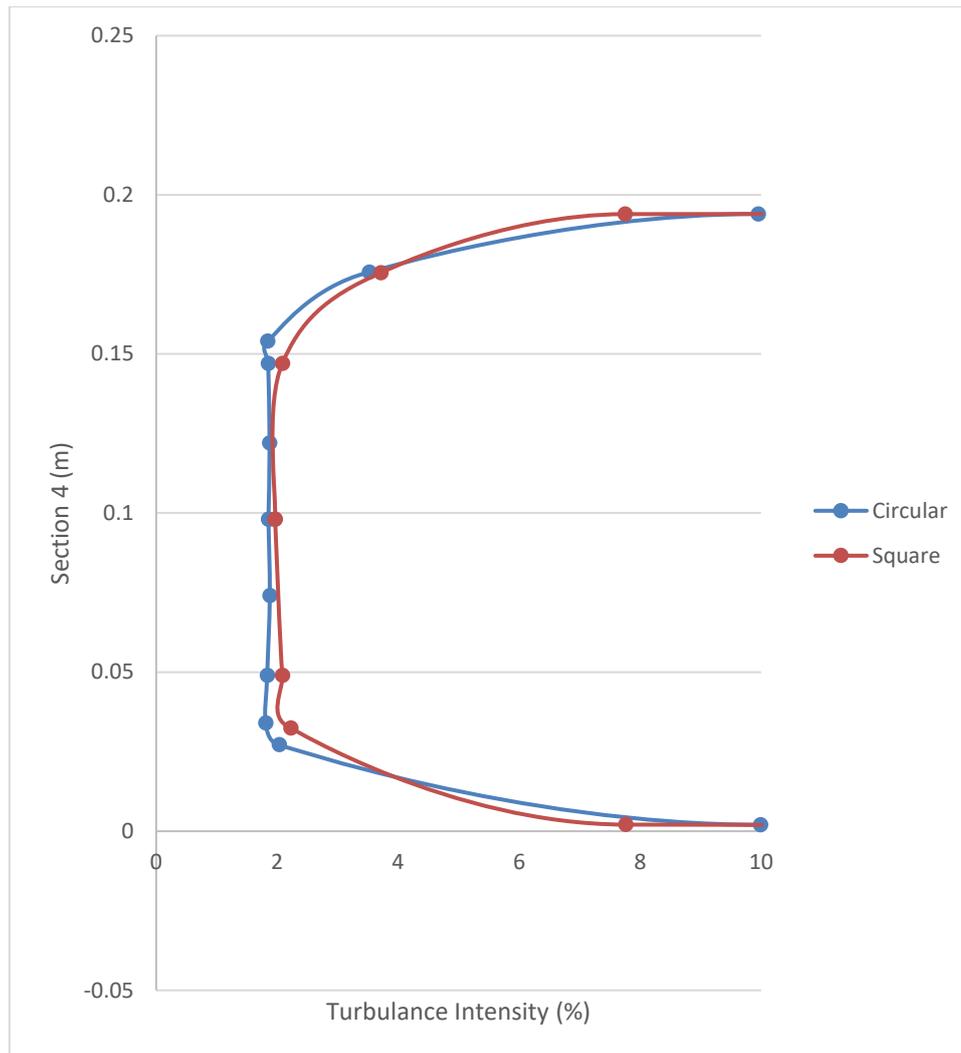


Figure 7. Turbulence intensity values at section 4

When the Figure 7 is examined, it is seen that when the turbulence intensity is compared, it has been determined that less turbulence occurs in the circular section wind tunnel compared to the square section wind tunnel.

4. CONCLUSION

Experimental studies are very important in designing a new product. In aerodynamic designs, wind tunnel tests are commonly used in experimental studies. In experimental studies, accuracy is the most critical point to verify the results. Turbulence intensity is the main drawback of wind tunnels. Hence in this study, turbulence intensity of the square and circular sectioned desktop size wind tunnel was investigated. Turbulence intensity distribution at section 4 in figure 5 showed that the circular sectioned wind tunnel has a better flow regime so it can give more accrued experimental results.

5. FUTURE STUDY

To verify the numerical analyses experimental study must be performed. In this study, desktop size wind tunnel models are used because in the future study they will be produced and experimentally investigated.

REFERENCES

- [1] Yannick D M, Jawahar H K, Szoke M, Ali S A S, Azarpeyvand M. Design and performance of an aeroacoustic wind tunnel facility at the University of Bristol. *Applied Acoustics* 2019; 155: 358–370.
- [2] Yang Y, Liu Y, Liu R, Shen C, Zhang P, Wei R, Liu X, Xu P. Design, validation, and benchmark tests of the aeroacoustic wind tunnel in SUSTech. *Applied Acoustics* 2021;175: 107847.
- [3] Martínez R M, Carpio A R, Pereira L T L, Herk S v, Avallone F, Ragni D, Kotsonis M. Aeroacoustic design and characterization of the 3D-printed, open-jet, anechoic wind tunnel of Delft University of Technology. *Applied Acoustics* 2020;170: 107504.
- [4] Niu W C, Ju Y L. System design and experimental verification of an internal insulation panel system for large-scale cryogenic wind tunnel. *Cryogenics* 2021;115: 103279.
- [5] Yia W, Zhoua P, Fanga Y, Guoa J, Zhonga S, Zhanga X, Huangb X, Zhouc G, Chen B. Design and characterization of a multifunctional low-speed anechoic wind tunnel at HKUST. *Aerospace Science and Technology* 2021;115: 106814.
- [6] Hoang N T B and Bui B V. Investigation of wind tunnel wall effect and wing-fuselage interference regarding the prediction of wing aerodynamics and its influence on the horizontal tail. *Journal of Mechanical Science and Technology* 2019; 33(6): 2737-2746.
- [7] Yi L, Changchuan X, Chao Y, Jialin C. Gust response analysis and wind tunnel test for a high-aspect ratio wing. *Chinese Journal of Aeronautics* 2016; 29(1): 91–103.
- [8] Koreanschi A, Gabor O S, Acotto J, Brianchon G, Portier G, Botez R M, Mamou M, Mebarki Y. Optimization and design of an aircraft's morphing wing-tip demonstrator for drag reduction at low speeds, Part II - Experimental validation using Infra-Red transition measurement from Wind Tunnel tests. *Chinese Journal of Aeronautics* 2017; 30(1): 164–174.
- [9] Joe M, Kammegne T, Botez R M, Grigorie L T, Mamou M, Mebarki Y. Proportional fuzzy feed-forward architecture control validation by wind tunnel

tests of a morphing wing. Chinese Journal of Aeronautics 2017; 30(2): 561–576.

[10] Zhang W, Sun J, Wang L, Wu J, He L. Rotor airfoil aerodynamic design method and wind tunnel test verification. Chinese Journal of Aeronautics 2020; 33(8): 2123–2132.

[11] Meng Y, Wan Z, Xie C, An C. Time-domain nonlinear aeroelastic analysis and wind tunnel test of a flexible wing using strain-based beam formulation. Chinese Journal of Aeronautics 2021; 34(1): 380–396.

[12] Oliviu , Gabora S, Koreanschia A, Boteza R M, Mamoub M, Mebarki Y. Numerical simulation and wind tunnel tests investigation and validation of a morphing wing-tip demonstrator aerodynamic performance. Aerospace Science and Technology 2016; 53: 136–153.

[13] Xua J, Fub Z, Baib J, Zhangc Y, Duand Z, Zhang Y. Study of boundary layer transition on supercritical natural laminar flow wing at high Reynolds number through wind tunnel experiment. Aerospace Science and Technology 2018; 80: 221–231.

[14] He S, Guo S, Liu Y, Luo W. Passive gust alleviation of a flying-wing aircraft by analysis and wind-tunnel test of a scaled model in dynamic similarity. Aerospace Science and Technology 2021; 113: 106689.

[15] Göv, I., Dogru M H, and Korkmaz U. "Improvement of the aerodynamic performance of NACA 4412 using the adjustable airfoil profile during the flight." Journal of The Faculty of Engineering and Architecture of Gazi University 34.2 (2019): 1110-1125.

[16] Doğru M H. "Investigation of Velocity Distribution and Turbulent Energy for the Different Tip Shaped Projectiles." Çukurova University Journal of the Faculty of Engineering and Architecture 32.3 (2017): 39-46.

[17] Göv İ, Doğru M H. Aerodynamic Optimization of Naca 0012 Airfoil. The International Journal of Energy and Engineering Sciences 2020; 5.2: 146-155.

[18] Ümit K, Göv İ, Doğru M H. Aerodynamic Analyses of Naca 63-215. The International Journal of Energy and Engineering Sciences 2020; 5.2: 156-166.

[19] Eyercioglu O, Gov K, Aksoy A. Abrasive Flow Machining of Asymmetric Spur Gear Forging Die. Harran University Journal of Engineering 2019; 4(1): 12-20.

[20] Göv K. Experimental investigation of the effects of the coolant on the performance parameters of electrical discharge drilling of some aerospace materials. Journal of the Faculty of Engineering and Architecture of Gazi University 2017; 32:1, 293-301.



[21] Gv K, Soydan O, Eyerciođlu . Improving the surface quality of Ti-6Al-4V alloy produced by electrical discharge machining with abrasive flow machining. Journal of the Faculty of Engineering and Architecture of Gazi University 2020; 35:3 1159-1170.